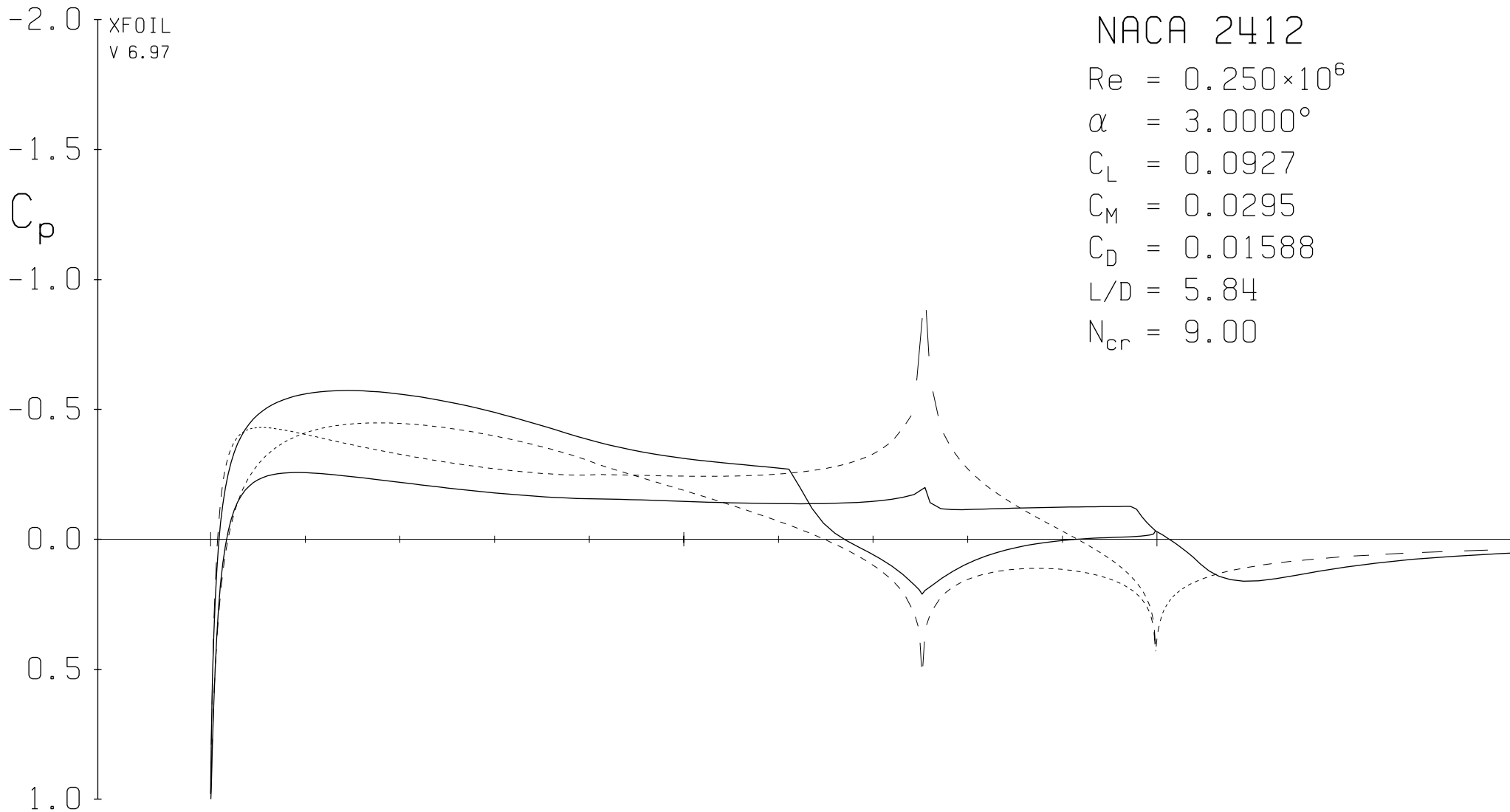


XFoil  
V 6.97



NACA 2412

$Re = 0.250 \times 10^6$

$\alpha = 3.0000^\circ$

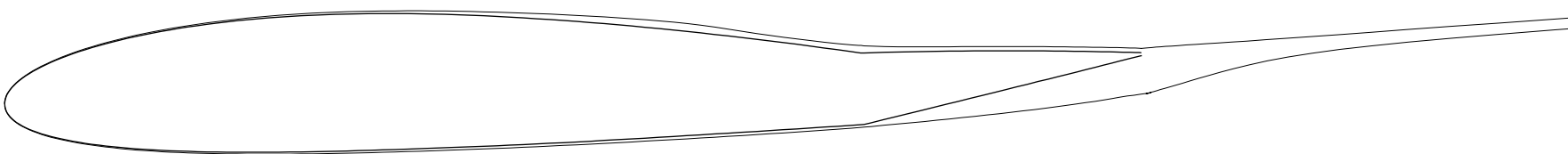
$C_L = 0.0927$

$C_M = 0.0295$

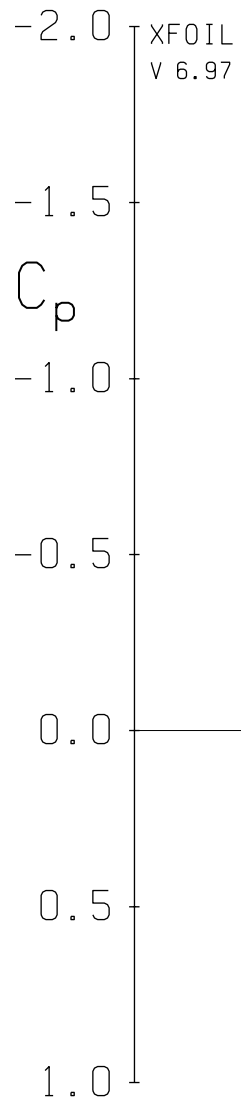
$C_D = 0.01588$

$L/D = 5.84$

$N_{cr} = 9.00$



$C_p$



XFOIL  
V 6.97

NACA 2412

$Re = 0.250 \times 10^6$

$\alpha = 3.0000^\circ$

$C_L = 1.0936$

$C_M = -0.1290$

$C_D = 0.01369$

$L/D = 79.90$

$N_{cr} = 9.00$

