



Analysis

Sequence

α

Start = -5.00

End = 8.00

Δ = 0.25

Store OpPoints

Store points outside the polar mesh

Init Calc

Analyze

Display

Half Wing

Lift Mom.

IDrag VDrag

TopTr BotTr

Cp Downw.

Anim Stream

Slow Fast

Axes Light

Surfaces Outline

Panels Vort.

X Y Z Reset

Lift and Moment Scale

Drag scale

Downwash scale

JW64" flap +3°

Wing span = 1600.50 mm

Wing area = 39.05 dm²

Plane weight = 1.35 kg

Wing load = 0.035 kg/dm²

Root chord = 332.00 mm

M.A.C. = 254.58 mm

Twist at tip = 0.0

Aspect Ratio = 6.6

Taper Ratio = 2.1

RT-tip sweep = -0.0

Point is out of the flight envelope

Qinf	=	13.9 m/s
Alpha	=	3.000°
Lift coef.	=	0.2866
Drag coef.	=	0.0097
Oswald	=	1.0159
Cl/cd	=	29.4800
Cm	=	-0.0199
Rolling Moment coef.	=	0.0000
Induced yawing moment coef.	=	-0.0000
Airfoil yawing moment coef.	=	0.0000
XCP	=	79.16 mm
Flap 1 Moment	=	0.001 N.m
Flap 2 Moment	=	0.001 N.m

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