



Analysis

Sequence

α

Start = -5.00

End = 8.00

$\Delta = 0.25$

Store OpPoints

Store points outside the polar mesh

Init Calc

Analyze

Display

Half Wing

Lift Mom.

IDrag VDrag

TopTr BotTr

Cp Downw.

Anim. Stream

Slow Fast

Axes Light

Surfaces Outline

Panels Vort.

X Y Z Iso Reset

Lift and Moment Scale

Drag scale

Downwash scale

JW64" flap 0°

Wing span = 1600.50 mm

Wing area = 39.05 dm²

Plane weight = 1.35 kg

Wing load = 0.035 kg/dm²

Root chord = 332.00 mm

M.A.C. = 254.58 mm

Twist at tip = 0.0

Aspect Ratio = 6.6

Taper Ratio = 2.1

RT-tip sweep = 0.0

QInf = 17.9 m/s

Alpha = 3.000°

Lift coef. = 0.1736

Drag coef. = 0.0108

Oswald = 1.0158

Cl/cd = 16.1206

Cm = 0.0199

Rolling Moment coef. = -0.0000

Induced yawing moment coef. = -0.0000

Airfoil yawing moment coef. = 0.0000

XCP = 32.45 mm

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