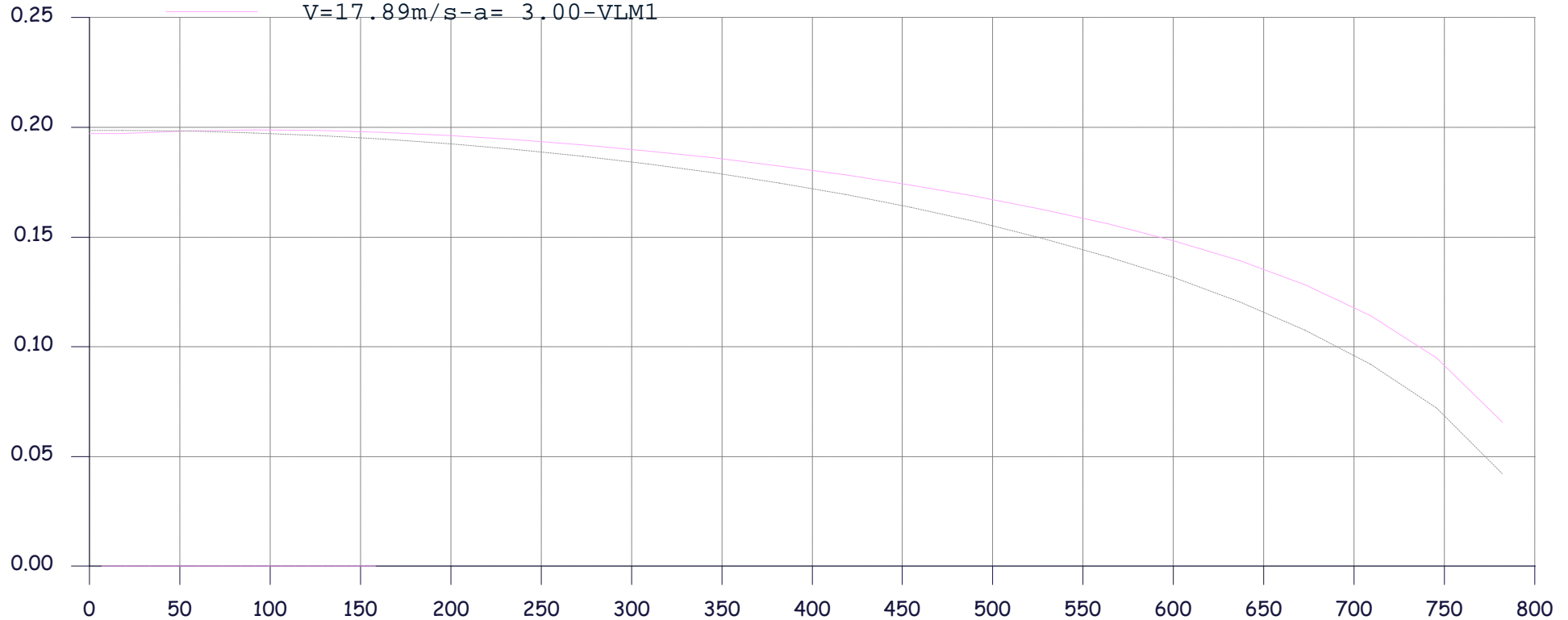


# Local lift

JW64" flap 0°\_Wing

V=17.89m/s-a= 3.00-VLM1



JW64" flap 0°\_Wing

Wing span = 1600.50 mm  
 Wing area = 39.05 dm<sup>2</sup>  
 Plane moment=refl.36 kg  
 Wing load = 0.035 kg/dm<sup>2</sup>  
 Root chord = 332.00 mm  
 M.A.C = 254.58 mm  
 Twist at tip = 0.0 °  
 Aspect Ratio = 6.6  
 Taper Ratio = 2.1  
 Rt-Tip Sweep = 0.0 °

— Top transition  
 — Bot transition  
 — Centre of Pressure

Pitching Moment coef. = 0.0199  
 Rolling Moment coef. = -0.0000  
 Airfoil Yawing Moment coef. = 0.0000  
 Induced Yawing Moment coef. = -0.0000

QInf = 17.9 m/s  
 Alpha = 3.000°  
 CL = 0.1736  
 VCd = 0.0093  
 ICd = 0.0014  
 Oswald = 1.0158  
 L/D = 16.1206  
 XCP = 0.0 mm

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