

```
QInf = 13.9 \text{ m/s}
JW64" flap +3° Wing
                                                                                               Alpha =
                                                                                                          3.000°
                                                                                                          0.2866
Wing span
             = 1600.50 \text{ mm}
Wing area
             = 39.05 \, dm^2
                                                                                                  Ved =
                                                                                                          0.0058
Plane wmoment=ref1.36ckgion
                                                                                                  ICd =
                                                                                                          0.0039
                                                                                              Oswald = 1.0159
Wing load
             = 0.035 \text{ kg/dm}^2
Root chord
             = 332.00 \text{ mm}
                                                                                                 L/D = 29.4800
                                                        Top transition
             = 254.58 \text{ mm}
M.A.C
                                                                                                  XCP =
                                                                                                          0.1 mm
                                                                              Pitching Moment coef. = -0.0199
Twist at tip =
                   0.0 °
                                                        Bot transition
Aspect Ratio =
                   6.6
                                                                               Rolling Moment coef. =
                                                                                                          0.0000
                                                        Centre of Pressirfoil Yawing Moment coef. =
Taper Ratio =
                   2.1
                                                                                                          0.0000
                  -0.00
                                                                        Induced Yawing Moment coef. = -0.0000
Rt-Tip Sweep =
                                                                                                          XFLR5 v3.21e
```

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